

5

## VARIABLE NOZZLE FOR A GAS TURBINE

The present invention relates to a nozzle for a gas turbine, which can be particularly applied to the first  
10 stage of a power turbine.

The present invention relates to a twin-shaft gas turbine and in particular, to a variable nozzle for a low pressure turbine.

Normally in twin-shaft turbines, the air pressurized  
15 by a compressor, is mixed with a combustible fluid and injected into a burner to generate hot combusted gases.

The latter flow through the nozzles of a high pressure turbine, which diverges them and accelerates them.

Downstream of the high pressure turbine, the gases  
20 then pass through a low pressure turbine, which extracts the remaining energy to feed a user.

Gas turbines for mechanical operations can have a fixed or variable nozzle, placed in the first stage of the low pressure turbine.

25 When using a variable nozzle, it is possible to ob-

tain a high operability of the turbine, at the same time maintaining the polluting emissions and efficiency of the turbine as constant as possible.

A fixed nozzle, on the other hand, is characterized  
5 by a higher aerodynamic efficiency accompanied however by a lower operability of the gas turbine.

For variable nozzles, there are clearances necessary for allowing its rotation.

A variable nozzle has two surfaces touched by hot  
10 combusted gases, opposite each other, of which one is pressurized and the other depressurized.

One of the disadvantages of a variable nozzle is that it has aerodynamic efficiency losses due to pressure drop losses of the flow of combusted gases through the  
15 clearances, accompanied by secondary losses arising from the latter, which are mainly due to the pressure differences between the pressurized surface and the depressurized surface.

An objective of the present invention is to provide  
20 a variable nozzle for a gas turbine, having improved performances which resemble those of a fixed nozzle, at the same time maintaining a high operability of the gas turbine with variations in its flow-rates.

Another objective of the present invention is to  
25 provide a reliable variable nozzle for a gas turbine.

These objectives according to the present invention are achieved by providing a variable nozzle for a gas turbine as illustrated in claim 1.

Further characteristics of the invention are indicated in the subsequent claims.

The characteristics and advantages of a variable nozzle for a gas turbine according to the present invention will appear more evident from the following, illustrative and non-limiting description, referring to the enclosed schematic drawings, in which:

figure 1 is a raised front view of a variable nozzle according to the present invention;

figure 2 is a raised sectional front view of the nozzle of figure 1 according to a line II-II passing through an upper end of the variable nozzle;

figure 3 is a raised sectional front view of the nozzle of figure 1, according to a line III-III passing through the intermediate part of the variable nozzle;

figure 4 is a raised sectional front view of the nozzle of figure 1 according to a line IV-IV passing through the hub of the variable nozzle;

figure 5 is a perspective view of the nozzle of figure 1;

figure 6 is a view from below of the nozzle of figure 1;

figure 7 is a raised side view of the nozzle of figure 1;

figure 8 is a view from above of the nozzle of figure 1;

5 figure 9 is a raised rear view from below of the nozzle of figure 1.

With reference to the figures, these show a variable nozzle 10 for a gas turbine fixed to a shaft 11 and capable of being rotated around its axis by means of activating means not shown in the figures.

The shaped variable nozzle 10 is suitable for minimizing pressure drops and consequently increasing the efficiency of the gas turbine.

Said variable nozzle 10 has a series of sections, preferably variable, substantially "C"-shaped, all facing the same direction, and preferably with the concavity facing upwards with respect to a base 90.

Each section of the series of sections represents a section of the variable nozzle 10 according to a surface having an axis parallel to the axis of the shaft 11.

Each section of the series of sections has a first rounded end 20 and a second rounded end 21.

The first end 20 of each section of the series of sections is situated along the axis of the shaft 11 according to an at least second degree curved line 60.

The series of sections is positioned along the axis of the shaft 11 and respectively defines two surfaces, an upper pressurized surface 12 and an opposite lower surface 14, which is depressurized, respectively, both  
5 touched by the hot combusted gases.

The pressure of the flow F of hot gas is exerted on the upper surface 12, whereas the opposite lower surface 14, is in depression.

The upper surface 12 is saddle-shaped and its saddle  
10 point corresponds to the intermediate section of the variable nozzle 10.

The upper surface 12, in a parallel direction to the axis of the shaft 11, is therefore convex, whereas in an orthogonal direction to said axis, it is concave, all the  
15 sections being substantially "C"-shaped.

The variable nozzle 10 has a first end portion 17, a second central portion 18, and a third hub portion 19.

The first portion 17 and the third portion respectively comprise an end section 30 and a hub section 50,  
20 which have minimum aerodynamic pressure drops which consequently improve the aerodynamic efficiency of the variable nozzle 10.

Furthermore, the pressure differences which are created between the upper pressurized surface 12 and the  
25 lower depressurized surface 14, always in respective cor-

respondence with said end section 30 and said hub section 50, are minimum and consequently the secondary aerodynamic losses are also minimum.

The forces which guide the flow of combusted gases through the clearances are thus reduced.

The second central portion 18, on the other hand, comprises the intermediate section 40.

There are no edge effects or secondary losses in correspondence with the second central portion 18, and consequently the aerodynamic efficiency in this portion of the variable nozzle 10 is greater.

For this reason, as there is a greater aerodynamic efficiency in the second central portion 18, the variable nozzle 10 is shaped so as to increase the aerodynamic charge thereon.

These results are also maintained with variations in the operating conditions of the gas turbine.

All of this is obtained by shaping the variable nozzle 10, positioning each section of the series of sections continuously one after another, and arranging the first end of each section of the series of sections in the direction of the axis of the shaft 11, along the at least second degree curved line 60.

Said curved line 60 lies on a surface 70 having an axis orthogonal to the axis of the shaft 11 and also

tilted with respect to the base 90 by an angle 80 different from  $0^\circ$  and lower than  $90^\circ$ .

Said curved line 60 is an at least second degree line and comprises a parabolic line or a hyperbolic line  
5 or a combination of these.

In a first preferred embodiment, said curved line 60 is preferably a parabolic line.

The variable nozzle 10 is therefore an arched nozzle, preferably parabolically arched.

10 In a second embodiment, said curved line 60 is preferably a hyperbolic line.

In a third embodiment, said curved line 60 is preferably a third degree line.

Said curved line 60, moreover, preferably has a  
15 maximum or minimum point.

It can thus be seen that a variable nozzle for a gas turbine according to the present invention achieves the objectives specified above.

Numerous modifications and variants can be applied  
20 to the variable nozzle for a gas turbine of the present invention, thus conceived, all included within the same inventive concept.

Furthermore, in practice, the materials used as also the dimensions and components, can vary according to  
25 technical demands.